1. Method for assembling into sets gas turbine engine components having flow passages comprising classifying the flow capability through the flow passages of each one of a plurality of the gas turbine engine components; and assembling into sets gas turbine engine components having the same flow capability classification.

2. The method of claim 1 wherein the engine components are gas turbine engine turbine blades with internal cooling passages and holes for film-cooling.

3. The method of claim 1 wherein the engine components are gas turbine engine turbine vanes with internal cooling passages and holes for film-cooling.

4. The method of claim 1 wherein the engine components are gas turbine engine turbine seals.

5. The method of claim 1 wherein the engine components are blades in any stage of the turbine section of a gas turbine engine.

6. The method of claim 1 wherein the engine components are vanes in any stage of the turbine section of a gas turbine engine.

7. The method of claim 1 wherein the engine components are seals in any stage of the turbine section of a gas turbine engine.

8. The method of claim 1 wherein the turbine components are blades in any stage of the compressor section of a gas turbine engine.

9. The method of claim 1 wherein the turbine components are vanes in any stage of the compressor section of a gas turbine engine.

10. The method of claim 1 wherein the turbine components are seals in any stage of the compressor section of a gas turbine engine.

11. The method of claim 1 wherein the flow classification includes a high-flow capability class and a low-flow capability class.

12. The method of claim 1 wherein the flow classification includes more than two flow capability classes ranging from a low-flow capability class to a high-flow capability class.

13. The method of claim 1 wherein the flow classification includes flow capability classes that are subsets of acceptable flow limits for gas turbine engine components having internal flow passages.

14. The method of claim 1 wherein the flow classification produces sets of components with increased high-temperature oxidation life capability.

15. The method of claim 1 wherein the flow classification produces sets of components with increased high-temperature creep life capability.

16. The method of claim 1 wherein the flow classification produces sets of components with increased high-temperature life capability.

17. The method of claim 1 wherein the required nominal amount of flow of the engine component can be reduced while maintaining high-temperature life capability.

18. The method of claim 1 wherein the required nominal amount of flow of the engine component can be reduced while maintaining the intended nominal performance of the component.

19. The method of claim 1 wherein the material of the engine component can be changed to a less capable material while maintaining the intended nominal performance of the component.

20. The method of claim 1 wherein the turbine inlet temperature is increased while maintaining the intended nominal performance of the component.

21. Gas turbine engine comprising gas turbine engine components assembled into sets, each said component having flow passages establishing a flow capability classification, wherein each said component in a set has the same flow capability classification.